# Airfoil Lab

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**Figure 1a.** Coefficient of lift on the y-axis versus angle of attack on the x-axis for the NACA0012 air foil at a chord Reynolds number of 1.5 \* 105. The experimental data has been plotted over published results using black square markers. Each marker is accompanied by error bars that represent the uncertainty of the data acquisition to within a 95% confidence interval.

A graph showing the value of a product

Description automatically generated with medium confidence

**Figure 1b.** Coefficient of drag on the y-axis versus angle of attack on the x-axis for the NACA0012 air foil at a chord Reynolds number of 1.5 \* 105. The experimental data has been plotted over published results using black square markers. Each marker is accompanied by error bars that represent the uncertainty of the data acquisition to within a 95% confidence interval.

A graph of a person

Description automatically generated with medium confidence

A graph of a number of points

Description automatically generated with medium confidence

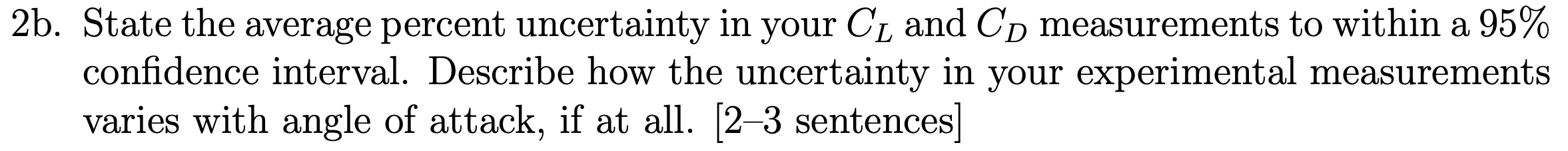
**Figure 1c.** (Left) Coeffcient of pressure on the y-axis versus distance along the chord on the x-axis for the NACA0012 airfoil at an angle of attack of 5 degrees at the chord Reynolds number of 1.5 \* 105. The blue markers represent the data measurements on the upper surface while the red markers represent the measurements on the bottom surface.(Right) Coeffcient of pressure on the y-axis versus distance along the chord on the x-axis for the NACA0012 airfoil at an angle of attack of 12 degrees at the chord Reynolds number of 1.5 \* 105. The blue markers represent the data measurements on the upper surface while the red markers represent the measurements on the bottom surface.

Short-Answer Questions

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2a. [insert your response here]



2b. [insert your response here]

A close-up of a paper

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2c. [insert your response here]

A close-up of a plane

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2d. [insert your response here]